

In the Claims:

1 1. (original) A lightweight, laminated structural component
2 made of thin metal plies comprising at least one sheet
3 metal component that is uninterrupted throughout its area
4 and at least one further sheet metal component constructed
5 as a framework forming a lattice, and a first adhesive bond
6 between said at least one sheet metal component and said
7 lattice.

1 2. (original) The lightweight, laminated structural component
2 of claim 1, wherein said at least one sheet metal component
3 comprises a first sheet metal ply that is uninterrupted
4 throughout its area, a second sheet metal ply that is also
5 uninterrupted throughout its area, a second adhesive bond
6 between said first and second uninterrupted metal plies to
7 form a first ply structure, and wherein said further sheet
8 metal component comprises a first sheet metal lattice, a
9 second sheet metal lattice and a third adhesive bond
10 between said first and second sheet metal lattices, to form
11 a second ply structure, and wherein said second ply
12 structure is bonded to said first ply structure by said
13 first adhesive bond.

1 3. (original) The lightweight, laminated structural component
2 of claim 1, further comprising stiffening members (18, 19)
3 operatively secured at least partly to said lattice for
4 forming a skin of an aircraft fuselage, said stiffening

members extending radially inwardly relative to a longitudinal central axis of said aircraft fuselage.

4. (original) The lightweight, laminated structural component of claim 3, wherein said stiffening members comprise stringers (18) extending in parallel to said longitudinal central axis, and ribs (19) extending circumferentially relative to said longitudinal central axis.

5. (original) The lightweight, laminated structural component of claim 3, wherein said lattice comprises struts (10, 11, 12) forming an integral part of said lattice and positioned for strengthening said lattice in accordance with load dependent criteria.

6. (original) The lightweight, laminated structural component of claim 5, wherein said struts (11, 12) extend in parallel to said stiffening members (18, 19) and/or at an angle relative to said stiffening members.

7. (original) The lightweight, laminated structural component of claim 1, wherein said lattice comprises sheet metal strip shaped lands positioned for facing into an aircraft fuselage, said strip shaped lands forming at least one sheet metal ply with open fields surrounded by said strip shaped lands.

1 8. (original) The lightweight, laminated structural component
2 of claim 1, wherein said at least one sheet metal component
3 and said further sheet metal component forming said lattice
4 have a thickness within the range of 0.5 mm to 5.0 mm.

1 9. (original) The lightweight, laminated structural component
2 of claim 1, wherein said at least one sheet metal component
3 and said at least one further sheet metal component are
4 made of a metal selected from the group of: alloys of
5 aluminum, alloys of titanium, steel alloys, alloys of
6 copper, alloys of zinc, and alloys of magnesium.

1 10. (currently amended) A method for manufacturing ~~[[a]]~~ the
2 lightweight, laminated structural component of claim 1,
3 comprising the following steps:

4 (a) preparing ~~[[a]]~~ said at least one sheet metal
5 component forming at least one sheet metal ply that is
6 uninterrupted throughout its area,

7 (b) preparing ~~[[a]]~~ said further sheet metal component
8 constructed as said framework ~~[[as a]]~~ forming said
9 lattice having strip shaped lands surrounding open
10 fields, and

11 (c) adhesively bonding said lattice to said at least one
12 sheet metal ~~[[ply-]]~~ ply to form said adhesive bond.

1 11. (original) The method of claim 10, wherein said adhesive
2 bonding is performed so that at least portions of said
3 lattice are adhesively bonded to said at least one sheet

4 metal ply and wherein said portions are determined by load
5 distribution patterns to which said structural component is
6 exposed.

1 12. (original) The method of claim 10, wherein said preparing
2 steps and said adhesive bonding step are performed as a
3 continuous, uninterrupted production operation.

1 13. (original) The method of claim 10, comprising using an
2 epoxy film as a bonding layer between said lattice and said
3 sheet metal component.

1 14. (original) The method of claim 10, further comprising
2 preparing at least two uninterrupted sheet metal plies,
3 adhesively bonding said at least two uninterrupted sheet
4 metal plies to each other, preparing at least one lattice,
5 and adhesively bonding said at least one lattice to said at
6 least two uninterrupted sheet metal plies.

1 15. (original) The method of claim 10, further comprising
2 securing stiffening members (18, 19) to said strip shaped
3 lands by any one or more of the following steps: adhesive
4 bonding, riveting and welding.

[RESPONSE CONTINUES ON NEXT PAGE]